

## FLADE GAS TURBINE ENGINE WITH FIXED GEOMETRY INLET

## ABSTRACT

An aircraft propulsion system includes a gas turbine engine having a fan section, at least one row of FLADE fan blades disposed radially outwardly of and drivingly connected to the fan section, the row of FLADE fan blades radially extending across a FLADE duct circumscribing the fan section, an engine inlet including a fan inlet to the fan section and an annular FLADE inlet to the FLADE duct. A fixed geometry inlet duct is in direct flow communication with the engine inlet. The fan section may include only a single direction of rotation fan or alternatively axially spaced apart first and second counter-rotatable fans in which the FLADE fan blades are drivingly connected to one of the first and second counter-rotatable fans. The row of FLADE fan blades may be disposed between rows of variable first and second FLADE vanes.